



# MACHAKOS UNIVERSITY

University Examinations 2021/2022 Academic Year

SCHOOL OF PURE AND APPLIED SCIENCES

DEPARTMENT OF MATHEMATICS AND STATISTICS

FOURTH YEAR FIRST SEMESTER EXAMINATION FOR

BACHELOR OF SCIENCE (MATHEMATICS AND COMPUTER SCIENCE)

SMA 434: GAS DYNAMICS

DATE: 2/9/2022

TIME: 8.30-10.30 AM

INSTRUCTION:

*Answer Question One and Any Other Two Questions*

## QUESTION ONE (30 MARKS)

- a) Describe how aerodynamics aids in the flying of airplanes (2 marks)
- b) A calorically perfect ideal gas is undergoing an isentropic process. What two factors can you consider to justify that the gas is indeed undergoing the process. (2 marks)
- c) Compute the density of air at  $8400 \text{ Nm}^{-2}$ ,  $560 \text{ K}$ .  
*Take the gas constant  $R=53.3 \text{ Jkg}^{-1} \text{ K}^{-1}$*  (3 marks)
- d) Consider a column vector  $h_i (i = 1, 2, 3, \dots, n)$  of  $n$  variables, considering two functions  $f_i(h_i)$  and  $g_i(h_i)$ , where all variables are functions of  $x$  and  $t$ , Formulate the necessary system of partial differential equation for the system to be in conservative form. 3 marks
- e) Consider a fluid flowing between a slot separated by two plates; the lower at  $y = 0$  is stationary and the upper at  $y = h$  is moving with a velocity  $U$ . Given that the velocity is only in the  $x$  direction and only a function of  $y$ , the flow has no imposed pressure gradient or body force and the flow has no acceleration, assuming constant viscosity and using the linear momentum principle
- i. Show that the flow is linear. (3 marks)

- ii. Suppose that the viscosity of the fluid is  $0.30 \text{ kgm}^{-1}\text{s}^{-1}$ . Compute the shear stress  $\tau$  in the fluid if it is moving with a velocity,  $U = 5 \text{ ms}^{-1}$  and the clearance  $h$  between the plates is 2.5 cm. (2 marks)
- f) Consider a gas flowing through a stream tube. Given that at one section of the tube, the flow has a pressure of  $200 \text{ Nm}^{-2}$ , a temperature of 300 K and is moving with a velocity of  $250 \text{ ms}^{-1}$ , Compute the temperature at another section of the tube where the pressure reduces to  $40 \text{ Nm}^{-2}$  and the speed increases to  $1100 \text{ ms}^{-1}$ . Take the specific heat of the gas at constant pressure,  $c_p = 2295 \text{ Jkg}^{-1}\text{K}^{-1}$ . (4 marks)
- g) Hydrogen gas has a static temperature of 298 K and stagnation temperature of 523 K. Calculate its Mach number and hence classify the disturbance as either subsonic or supersonic. Take the ratio of specific heats to be  $\gamma = 1.41$ . (3 marks)
- h) Consider air at a temperature of 600 K flowing through a converging-diverging nozzle which has a cross sectional area of  $1 \text{ cm}^2$ . Given that the initial mass flow rate is  $0.04125 \text{ kgs}^{-1}$ , the initial and exit pressures are 200 kPa and 191.5 kPa respectively. Compute;
- i. Mass flow rate at the exit, (3 marks)
- ii. Exit velocity. (5 marks)
- Take  $\gamma = 1.4$ ,  $R = 287 \text{ Jkg}^{-1}\text{K}^{-1}$  and  $c_p = 1004.5 \text{ Jkg}^{-1}\text{K}^{-1}$ .

### QUESTION TWO (20 MARKS)

- a) Consider a flow between a slot separated by two plates; the lower at  $y = 0$  is stationary and the upper at  $y = h$  is moving with a velocity  $U$ . Given that the velocity is only in the  $x$  direction and only a function of  $y$ , the flow is driven by a pressure difference in the  $x$  direction such that at  $x = 0, P = P_0$  and at  $x = l, P = P_1$ . If there is no body force, the flow has no acceleration, assuming constant viscosity and using the linear momentum principle, Compute the velocity profile parameterized by  $P_0, P, h, U$  and  $\mu$ . (10 marks)
- b) Considering an ideal gas where pressure  $P$  is a function of temperature,  $T$  and Volume,  $V$  given by  $P(T, V) = \frac{RT}{V}$ . Using the laws of thermodynamics, derive the relation for;
- i. The total internal energy  $e(T) = e_0 + c_v(T - T_0)$  (4 marks)
- ii. The gas constant  $R = c_p - c_v$  (3 marks)

c) Given that the speed of sound in a general material,  $c(T, \rho) = \sqrt{\left(\frac{dp}{d\rho}\right)_T + \frac{T}{c_v \rho^2} \left[\left(\frac{dP}{dT}\right)_\rho\right]^2}$

Simplify this expression and prove that for an ideal gas, the speed is only a function of temperature and given by  $c(T) = \sqrt{\gamma RT}$  (3 marks)

### QUESTION THREE (20 MARKS)

a) Consider a calorically perfect ideal gas flowing through a duct of length 150m having a uniform cross section of area  $0.03 \text{ m}^2$ . Given that at one end of the duct, the gas has a pressure of 150 kPa, a temperature of 300 K and is flowing at a velocity of  $12 \text{ ms}^{-1}$ . If the gas is isobarically heated with a constant heat flux along the entire surface of the duct such that at the other end it has a pressure of 150 kPa and a temperature of 500K, Calculate;

i. The mass flow rate, (4 marks)

ii. Velocity of the gas at the other end. (4 marks)

Take  $R=287 \text{ Jkg}^{-1}\text{K}^{-1}$

b) Consider an airplane flying through still air at a velocity of  $200 \text{ ms}^{-1}$ . Assuming steady isentropic flow of a calorically perfect ideal gas, given that the ambient air has a temperature of 288 K and a pressure of 101.3 kPa, Compute;

i. The Mach number and hence classify the disturbance, (4 marks)

ii. Temperature, pressure and density at the nose of the airplane. (8 marks)

Take  $\gamma=\frac{7}{5}$  and  $R=287 \text{ Jkg}^{-1}\text{K}^{-1}$

### QUESTION FOUR (20 MARKS)

Consider air flow through a normal shock wave. Given that the upstream conditions are  $u_1 = 600 \text{ ms}^{-1}$ ,  $T_{01} = 500 \text{ K}$ ,  $P_{01} = 700 \text{ kPa}$ , using Rankine-Hugoniot relations Calculate the downstream conditions  $M_2$ ,  $u_2$ ,  $T_2$  and  $P_{02}$ . Take  $\gamma=1.4$ ,  $c_p = 1000 \text{ Jkg}^{-1}\text{K}^{-1}$  and  $R=287 \text{ Jkg}^{-1}\text{K}^{-1}$

### QUESTION FIVE (20 MARKS)

a) Consider a gas flowing over a wedge inclined at an angle of  $20^\circ$  to the horizontal. For a calorically perfect ideal gas, if the upstream conditions are  $P_1 = 100 \text{ kPa}$ ,  $T_1 = 300 \text{ K}$  and  $M_1 = 3$ . Using Hodograph and characteristic relations, compute the downstream pressure and temperature for a shock angle of;

- i.  $\beta = 37.76^\circ$  relating to a weak oblique shock wave, (5 marks)
- ii.  $\beta = 82.15^\circ$  relating to a strong oblique shock wave, (5 marks)
- iii.  $\beta = -9.91^\circ$  relating to a rarefaction shock. (5 marks)
- b) Consider a calorically perfect ideal gas. Given that it has a temperature of 300 K and is moving with a velocity of  $500 \text{ ms}^{-1}$ , compute the Prandtl-Mayer function  $\nu(M)$ .  
*Take  $\gamma = 1.4$  and  $R = 287 \text{ Jkg}^{-1}\text{K}^{-1}$*  (5 marks)